

Delamination Model Using Damage Mechanics Applied to New Composite for Orthopaedic Use

Djebbara Benzerga^{1,*}, Abdelkader Haddi², Antoine Lavie²

¹Université des Sciences et de la Technologie Mohamed Boudiaf, Laboratoire Structures de Composite et Matériaux Innovants (LSCMI), Faculté de Génie, BP 1505 Oran El M'Naouer 31000, Algeria

²Univ Lille Nord de France, F-59000 Lille, France, Laboratoire Génie Civil et géo-Environnement (LGCgE), EA 4515, Faculté des Sciences Appliquées, Université d'ArtoisF-62400 Béthune, France

Abstract In this study, delamination initiation and propagation are numerically predicted of a new laminated composite reinforced with natural organic load. This new composite is obtained from a laminated composite woven by incorporating a natural non-polluting organic load (granulates of date cores) which becomes hybrid composite. The new economical hybrid composite material is made of an organic matrix containing methyl methacrylate, a woven reinforcement including a reinforcing glass fiber and a fabric perlon having an absorbing role. The walk cycle has been used to determine the operating conditions of tibiae prosthesis. Hence, the deflection tests were validated by orthopedist experts. This hybrid composite is used for the manufacture of composite for orthopaedic use and also for other applications in the future. The bonding conditions between layers are characterised by jumps in displacements which are proportional to the traction stresses. In order to describe the interface damage, an approach based on damage mechanics considering a softening stress-relative displacement law is presented. The objective is to develop a delamination model that can predict delamination growth in the new woven hybrid composite used in tibiae prosthesis. 3ENF tests were carried out on the new woven composite to detect delamination phenomenon. We assume that the interface has a bi-linear softening behaviour and regarded as being a whole of several interfacial bonds. Each bond is supposed to be made up of three stiffnesses acting in the three delamination mode directions. Numerical simulations are compared with experimental results carried out in single mode delamination tests, in the mixed-mode bending test to validate the delamination model and in 3ENF test of a new woven laminated composite for orthopaedic use. A scalar damage variable is introduced and the degradation of the interface stiffness is established. The damage model is implemented into a commercial finite element ANSYS to simulate delamination in mode II. Numerical results on (90, 45₂, 0) are in good agreement with experimental observations.

Keywords Composites materials, Interface, Damage, Delamination

1. Introduction

Delamination is an important phenomenon of damage in laminated composite materials due to weakness of reinforcement through the thickness. The study of the delamination of a laminate may be performed using an approach of fracture mechanics or by introducing appropriate constitutive laws of the interface between the layers constituting the laminate. From a physical point of view, it is reasonable to assume that the second approach can be related to fracture mechanics. In fact, when decohesion occurs between adjacent layers, there is evolution of delaminated surface which is equivalent to the propagation of a crack in a direction a priori known. The literature dealing

with the phenomenon of delamination is very large. A presentation of several structures subjected to the phenomenon of delamination, can be found [1-2]. The delamination phenomenon can be caused by concentration of interlaminar stresses that occur in the vicinity of the free edges or in around of the holes in laminated plates [3]. In addition, the interlaminar defects can grow under a compressive loading. In this case the thin laminated layers degrade (debonding interfaces) and are responsible for increased stresses in the vicinity of the boundaries of delaminated surfaces. In the analysis of the delamination can be distinguished the stage of the crack initiation from the phase of the crack propagation. For prediction of the initiation of a crack from a free edge, the technical calculations of the effect of the edge of elasticity [4-5], related to criteria based on the average of the normal stresses on a characteristic distance from the edge of the structure [6] are usually used in post-processor of an elastic design in laminated structures. Delamination does not occur

* Corresponding author:

djeb_benz@yahoo.fr (Djebbara Benzerga)

Published online at <http://journal.sapub.org/ijme>

Copyright © 2014 Scientific & Academic Publishing. All Rights Reserved

necessarily where the stresses are highest. In the phase of the propagation of a delamination established, approaches based on the linear fracture mechanics are generally used. The rate of energy release G is a parameter that is often used to describe the behavior of the phenomenon of delamination in composite materials and structures. G is defined as the energy released from the newly fractured surface and compared to the critical value G_c (This method is used by many authors for the study of crack propagation [7-8]), but not treat the problem of the initiation of a delamination crack. In contrary the approach of the damage mechanics of the composite can describe the initiation of delamination. The rate of energy release is calculated from the forces and nodal displacements [9-10]. The state equations and the evolution laws of the interface provided in the context of thermodynamics are described in [11-17]. Models of elastic and damageable interface are presented in [18-22] and generalized in [23]. In these models, special interface elements are applied in areas where the delamination phenomenon is likely to occur. Elements plane strain with cubic interpolation functions were introduced for discretization of the laminate ply [24]. Other models have been developed for modeling damage layer [25] and interface phenomena [26]. These models are based on the damage mechanics. The interface is considered as a three-dimensional medium with negligible thickness compared to the other dimensions. Therefore, the interface can be considered as two-dimensional entity which transfers traction and displacement from one layer to the other [27]. The interface is assumed to be dependent on the fibre orientation of adjacent layers and it is assumed to be elastic and damageable. Delamination may be caused by interlaminar stress [4]. The objective of this paper is to present a method to simulate progressive delamination based on a new mixed-mode failure criterion in the context of damage mechanics. This study will highlight the positive contribution of the powder core dates incorporated in the new woven composite. The date cores powder incorporation has an increase effect of the mechanical characteristics giving to the hybrid composite a better behaviour and reducing certain types of degradation like delamination.

2. Delamination Analysis

It is widely recognised that the major contribution to delamination fracture resistance is given by the damage developing in matrix-rich interlaminar layer. Delamination is created by an important accumulation of cracks in the matrix. For this reason the delamination occurs in general later in the history of the laminate damage. Transverse matrix cracking, when it is propagated, can reach the interface between two layers of different fibre orientation. The interface between two adjacent layers can debond under interlaminar tension stresses σ_{33} and/or shear stresses σ_{13} and σ_{23} (figure 1).

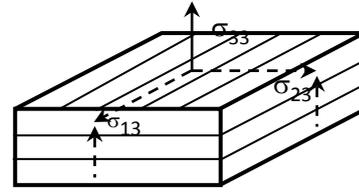


Figure 1. Interlaminar stresses which cause delamination

The mesomodelling of laminates is used. It can describe the degradation of the structure until its ultimate rupture [28]. At the meso-scale, a laminate is modelled as a stacking sequence of layers and damageable elastic interlaminar interfaces. The interface is considered as a two-dimensional entity which ensures traction and displacement transfer from one ply to another (only stress transfer in the damageable case) [29] (figure 2). This modeling coupled with the damage mechanics involves the phenomena of delamination (initiation and propagation). The jump of displacement in the orthotropic base of the interface is [30]

$$U = U^{\text{sup}} - U^{\text{inf}} = \begin{Bmatrix} u_{13} \\ u_{23} \\ u_{33} \end{Bmatrix}_{\text{sup}} - \begin{Bmatrix} u_{13} \\ u_{23} \\ u_{33} \end{Bmatrix}_{\text{inf}} \quad (1)$$

Where U is the vector of relative displacements between two homologous points, which can be obtained from the displacements fields of the upper and lower surface.

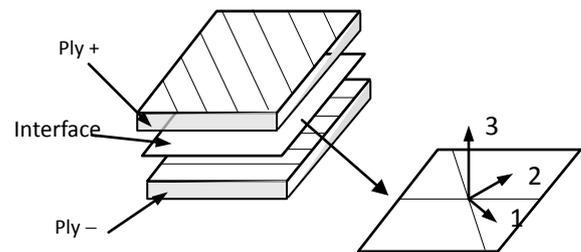


Figure 2. Interface: surface entity between two adjacent layers

For damage caused in tension and shear through the thickness of the shell, the interface damage modelling is applied. This model allows the analysis of tests in fracture mechanics [31], from the method of determination of the parameters of the model. The mechanical behaviour of the interface depends only on the relative angle between the adjacent plies [32].

The behaviour of the interface is assumed to be dependent on the fibre orientation of adjacent layers. This dependence can be taken into account by introducing a local reference frame of 'natural' orthotropic directions 1, 2 and 3; 3 is normal to the interface, 1 and 2 are the bisector of the angle formed by the fibres of the upper and lower layers (Figure 2) [33]. The constitutive equations between the interlaminar stresses and the displacement jumps in the orthotropic base

of the interface are

$$e_D = S_D \sigma_D \quad (2)$$

with

$$e_D = (2e_{13}, 2e_{23}, e_{33})$$

$$\text{and } \sigma_D = (2\sigma_{13}, 2\sigma_{23}, \sigma_{33})$$

S_D the constitutive matrix given in inverse form through the thickness such as

$$s_D = \begin{pmatrix} 1/G_{13}(1-d_{13}) & 0 & 0 \\ 0 & 1/G_{23}(1-d_{23}) & 0 \\ 0 & 0 & 1/E_{33}(1-d_{33}) \end{pmatrix} \quad (3)$$

where E_{33} is the Young modulus through the thickness G_{13} et G_{23} are the shear modulus in direction 1 and 2. The damage parameters d_{13}, d_{23}, d_{33} , represent the shear and Young modulus reduction due to the interface damage. Delamination occurs in the thin layers between the plies of the composite. In delamination models, the stresses are assumed to be constant through the thickness t of the interface. By considering the classical relations deformation-displacements, the components of deformations in the thickness can be approached by

$$e_{13} = \frac{1}{2} \frac{u_{13}}{t}, e_{23} = \frac{1}{2} \frac{u_{23}}{t}, e_{33} = \frac{u_{33}}{t} \quad (4)$$

Substituting equations (2) and (3), in the relation stress-strains of the interface, one obtains the relations stresses-displacements within the limit $t \rightarrow 0$

$$\begin{aligned} \sigma_{13} &= k_{13}(1-d_{13}) \\ \sigma_{23} &= k_{23}(1-d_{23}) \\ \sigma_{33} &= k_{33}(1-d_{33}) \end{aligned} \quad (5)$$

Before delamination initiation these relations are written quite simply

$$\sigma_{13} = k_{13}u_{13}, \sigma_{23} = k_{23}u_{23}, \sigma_{33} = k_{33}u_{33} \quad (6)$$

with $k_{13} = G_{13}/t, k_{23} = G_{23}/t, k_{33} = E_{33}/t$

The relations (5) are the constitutive equations of an elastic and damageable interface. They link the normal and shear jumps through the interface to the stresses. The parameters of the interface damage d_{13}, d_{23}, d_{33} , are three scalar damage variables which vary between 0 (no damage) and 1 (totally damaged).

3. Interface Damage Model

It is assumed that the mechanical behaviour of the interface ($\sigma_{i3} - u_{i3}$) follows the law described in figure 3 where $u_{i3,0}$ and $u_{i3,m}$ ($i=1,2,3$) correspond to the displacements obtained for the maximum stress $\sigma_{i3,m}$ at the final rupture of the interface.

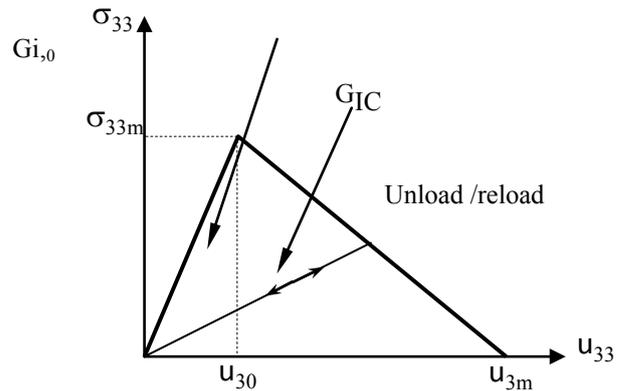


Figure 3. Relation stress-relative displacement $\sigma_{i3} - u_{i3}$ of the interface

The proposed damage model is able to describe three possible modes of delamination (figure 4) and is based on an indirect use of rupture mechanics [34]. The model is completed by evolution equations for the interface damage parameters, which may be derived using the CDM framework described in [35]. Damage evolution is sometimes made a function of the damage energy release rate divided by a critical value [36]. The relative displacement and the damage energy release rate are related in a trivial mode. Hence in the present formulation, damage evolution is made a function of the relative displacement within a unit volume of material. The evolution of damage follows a simple bilinear relationship given by [37]

$$d_{i3} = \frac{u_{i3,m}}{(u_{i3,m} - u_{i3,0})} \left[1 - \frac{u_{i3,0}}{u_{i3}} \right] \quad (8)$$

with $u_{i3,0} = \sigma_{i3,m} / k_{i3}^0$

$$u_{i3,m} = 2G_{i,C} / \sigma_{i3,m} \quad (9 \text{ a, b})$$

where i can be used to represent mode I, II or III, and $u_{i3,m}$ is the strain at zero stress or damage = 1 (propagation), and $u_{i3,0}$ is the relative crack opening displacement at maximum stress or damage = 0 (initiation).

According to [38], the function stress-displacement (5) has a triangular form (figure 3). The law of the damage evolution contains two parameters $\sigma_{i3,m}$ and $G_{i,C}$

which is the critical energy of rupture or the work provided per unit of area for debonding of the interface. The interface initially elastic degrades gradually when the stress reached the value $\sigma_{i3,m}$ so that the energy of rupture in any mode is entirely absorbed at the separation [39, 17]. The interface is assumed to be made of several interfacial bonds and each bond is supposed to be constituted of three spring elements. Each element acts in one direction.

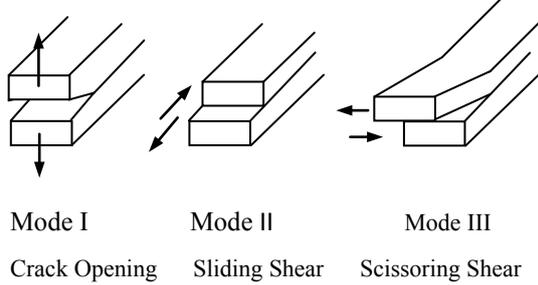


Figure 4. Modes of delamination fracture

The critical energies of rupture G_{IC} , G_{IIC} and G_{IIIC} can be calculated as:

$$\begin{aligned} G_{IC} &= \int_0^{u_{33,m}} \sigma_{33} du_{33}, \\ G_{IIC} &= \int_0^{u_{13,m}} \sigma_{13} du_{13} \\ G_{IIIC} &= \int_0^{u_{23,m}} \sigma_{23} du_{33} \end{aligned} \quad (10)$$

It can be shown that the area under the curve in figure 3 is equal to the fracture energy G_{iC} ($i = I, II, III$).

In mixed mode of delamination analysis it is necessary to include simultaneously the three damage modes. To begin with, consider the stored energy function [41]:

$$\begin{aligned} 2\zeta_d &= (1 - d_{13}) k_{13}^0 [u_{13}]^2 \\ &+ (1 - d_{23}) k_{23}^0 [u_{23}]^2 + (1 - d_{33}) k_{33}^0 [u_{33}]^2 \end{aligned} \quad (11)$$

Considering an isotropic damage $d = d_{13} = d_{23} = d_{33}$, (the interface is a medium of very small thickness), Eq. (11) becomes

$$2\zeta_d = (1 - d) \left[k_{13}^0 [u_{13}]^2 + k_{23}^0 [u_{23}]^2 + k_{33}^0 [u_{33}]^2 \right] \quad (12)$$

where $d \in [0; 1]$ is the scalar damage variable.

The work-conjugate of the damage variable follows from the classical thermodynamic argument as [42]:

$$\begin{aligned} Y &= -\frac{\partial \zeta_d}{\partial d} \\ &= \frac{1}{2} \left[k_{13}^0 [u_{13}]^2 + k_{23}^0 [u_{23}]^2 + k_{33}^0 [u_{33}]^2 \right] \\ &= Y_{13} + Y_{23} + Y_{33} \end{aligned} \quad (13)$$

Using the same penalty stiffness in mode II and III ($k_{13}^0 = k_{23}^0 = k_{II}^0$), and assuming the delamination mechanisms in mode II and mode III to be same. Therefore, Mode III can be combined with Mode II by using a total tangential displacement u_{II} defined as the norm of the two orthogonal tangential relative displacements u_{13} and u_{23} as

$$u_{II} = \sqrt{u_{13}^2 + u_{23}^2} \quad (14)$$

Based on (13), the mixed-mode energy release rate Y can be expressed as:

$$\begin{aligned} Y &= \frac{1}{2} k_{33}^0 \left[u_{33}^2 + \gamma^2 u_{II}^2 \right] \\ \text{with } \gamma &= \left(\frac{k_{II}^0}{k_3^0} \right)^{1/2} \end{aligned} \quad (15)$$

A mode mixity parameter β_1 can thus be defined as:

$$\beta_1 = \gamma \tan(\psi) \quad (16)$$

ψ being the loading angle:

$$\psi = \arctan\left(\frac{u_{II}}{u_{33}}\right) \in [0, \pi/2] \quad (17)$$

where by the expressions of the pure-mode contributions to (13) follow as:

$$Y_{33} = \frac{1}{1 + \beta_1^2}; \quad Y_{II} = \frac{\beta^2}{1 + \beta_1^2} \quad (18)$$

In particular, assuming that initiation of damage can be predicted using a Hashin-type criterion [20], i.e.

$$\left(\frac{Y_{33}}{G_{0I}} \right)^{\alpha_1} + \left(\frac{Y_{II}}{G_{0II}} \right)^{\alpha_1} = 1 \quad (19)$$

where G_{0I} and G_{0II} are the initial damage thresholds for a given loading angle (17) while α_1 is positive model parameter. The initial mixed-mode threshold Y_0 is computed from (19) as:

$$Y_0 = \frac{(1 + \beta_1^2) G_{0I} G_{0II}}{\left[(G_{0II})^{\alpha_1} + (\beta_1^2 G_{0I})^{\alpha_1} \right]^{1/\alpha_1}} \quad (20)$$

whence the pure-mode threshold energies G_{0I} and G_{0II} (fig.3) are recovered in the limit as $\psi \rightarrow 0$ and $\psi \rightarrow \pi/2$ respectively.

The criteria used to predict delamination propagation

under mixed-mode loading conditions is generally established in terms of the energy release rates and fracture toughness. We use two criteria to predict the interaction of energy release rates in mixed mode. The first criterion is a power law [43], the second is B-K criterion [45], recently developed by Benzeggagh and Kenane.

$$\left(\frac{G_{33}}{G_{IC}}\right)^\lambda + \left(\frac{G_{II}}{G_{IIC}}\right)^\lambda = 1 \quad (21)$$

$$G_T = G_{IC} + (G_{IIC} - G_{IC}) \left(\frac{G_{IIC}}{G_T}\right)^\eta \quad (22)$$

with $G_T = G_I + G_{II}$ and η a parameter.

The energies released at failure are computed from (fig. 3):

$$G_{XC} = \int_0^{u_{i3,m}} \sigma_{i3} du_{i3} \quad (23)$$

avec $X = I, II, III$ et $i = 1, 2, 3$

The propagation of decohesion takes place for:

$$Y_f = \begin{cases} \frac{(1 + \beta_1^2) G_{IC} G_{IIC}}{\left[(G_{IIC})^{\alpha_1} + (\beta_1^2 G_{IC})^{\alpha_1} \right]^{\gamma_{a1}}} \text{ for law (21)} \\ G_{IC} + (G_{IIC} - G_{IC}) \left(\frac{\beta_1^2}{1 + \beta_1^2} \right)^\eta \text{ for law (22)} \end{cases} \quad (24)$$

In order to account for irreversibility, the maximum over time value of the mixed-mode energy release rate $Y(t)$ is defined as, at time τ

$$Y_{\max}(t) = \max_{t \leq \tau} \{Y\} \quad (25)$$

The constitutive law (5) could be expressed as

$$\sigma_{i3} = (1 - d) k u_{i3} \quad (26)$$

In order to avoid interpenetration for compression situations, simple contact logic already available in most FE codes could be used. Instead, the following condition is added to Eq. (5):

$$\sigma_{33} = k_{33} u_{33} \iff u_{33} < 0 \quad (27)$$

Where only one damage variable is used, and computed as

$$d = \frac{\sqrt{Y_f}}{\sqrt{Y_f} - \sqrt{Y_0}} \left(1 - \frac{\sqrt{Y_0}}{\sqrt{Y_{\max}}} \right) \quad (28)$$

(if $d < 1$; $d = 1$ otherwise)

4. Element Formulation

In this study the delamination has been modeled by the spring elements, *COMBIN14*. We assume that there is no friction between the lips of the crack (perfect sliding case) [46]. For each position on the crack front of the initial interface crack, the damage is calculated and compared with the critical value ($d = 1$). When the damage is bigger the crack grows one step at the evaluated position and the spring element is disabled at this location. These ‘‘spring’’ elements, used for the elastic interface, have no thickness. This satisfies the condition of very thin interfacial zone comparatively to the dimensions of the constituents. For the numerical modeling the elastic interface was represented by a spring layer which resists normal extension and shear deformation (Fig. 5). A subroutine was developed and implemented into the FE code to model delamination growth simulation.

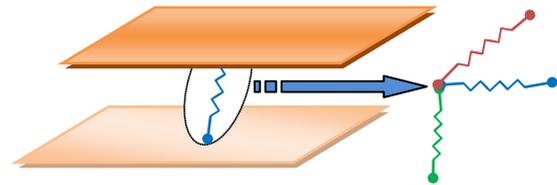


Figure 5. Rheological representation of the interfacial bond based on the association of three springs

5. Numerical Simulations

Three test problems were selected to validate the proposed modelling. The first problem consists of the double cantilever beam (DCB) test used to determine Mode I toughness. Figure 6a shows displacement loading. The second problem modelled consists of the end-notched flexure test (ENF) used for Mode II toughness. The third test is the mixed bending mode test (MMB) with $G_{II} / (G_I + G_{II})$ equal to 50%. All three of these problems have analytical solutions that were developed by Mi and Crisfield [47]. The geometry of test specimen composed of two plies is shown in table I and the properties of the composite in table II. Figures 6b, 7 and 8 show the numerical predictions and the experimental or analytical data for all the test cases simulated. A plot of reaction force as a function of the applied end displacement d is shown in Figure 6 for DCB tests. The load-deflection responses for the finite element model and the analytical prediction for 3ENF tests are shown in Figure 8. It can be observed that the comparisons of numerical simulations with experimental results are quite good. The load deflection curve calculated from FEM and beam solutions are shown in Figure 7. The beam solution assumes a linear interaction between the energy release rates. Reasonably good correlation is obtained among all three methods.

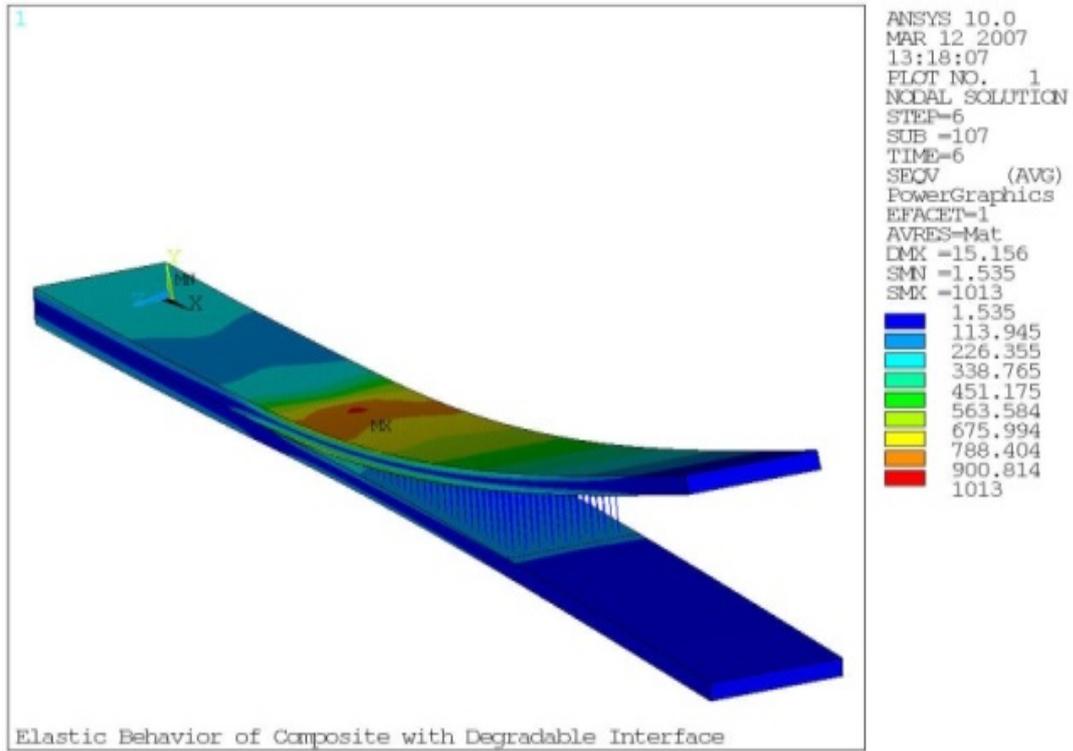


Figure 6a. DCB displacement loading

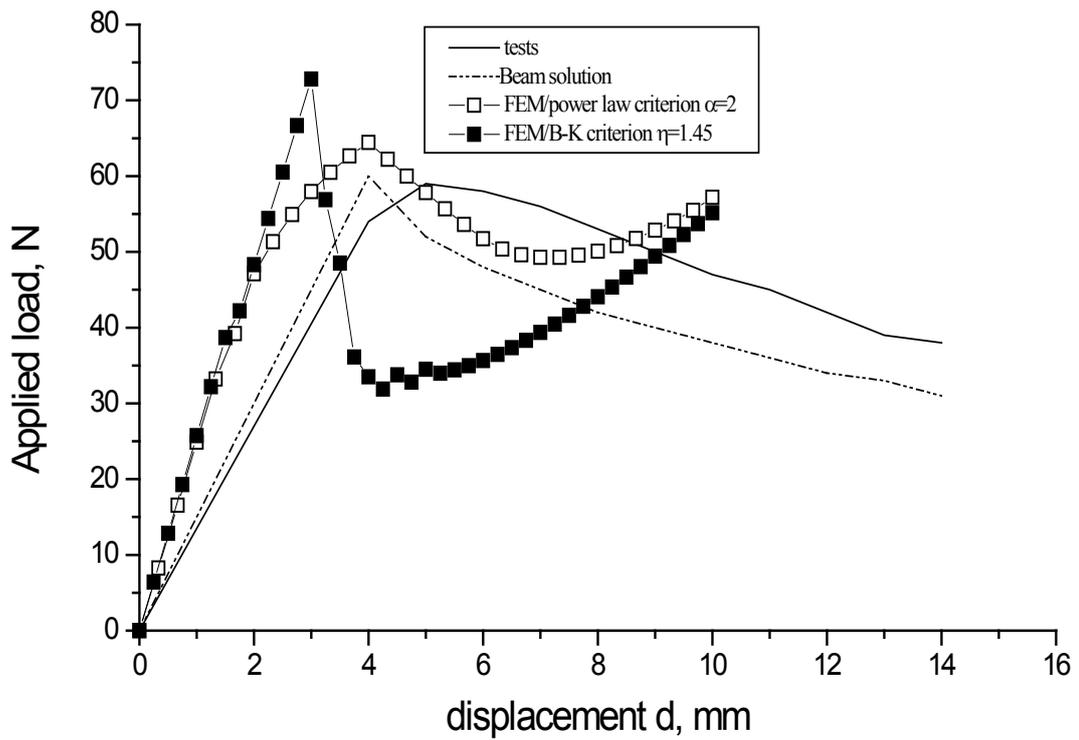


Figure 6b. Load-deflection response of DCB test

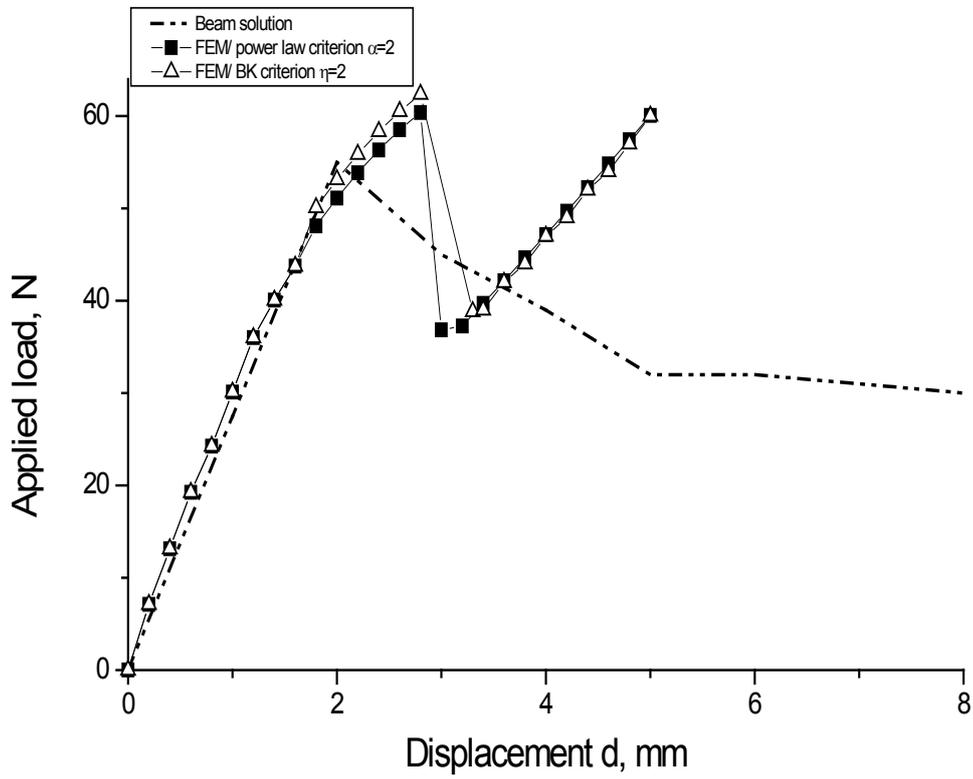


Figure 7. Load-deflection response of MMB tests

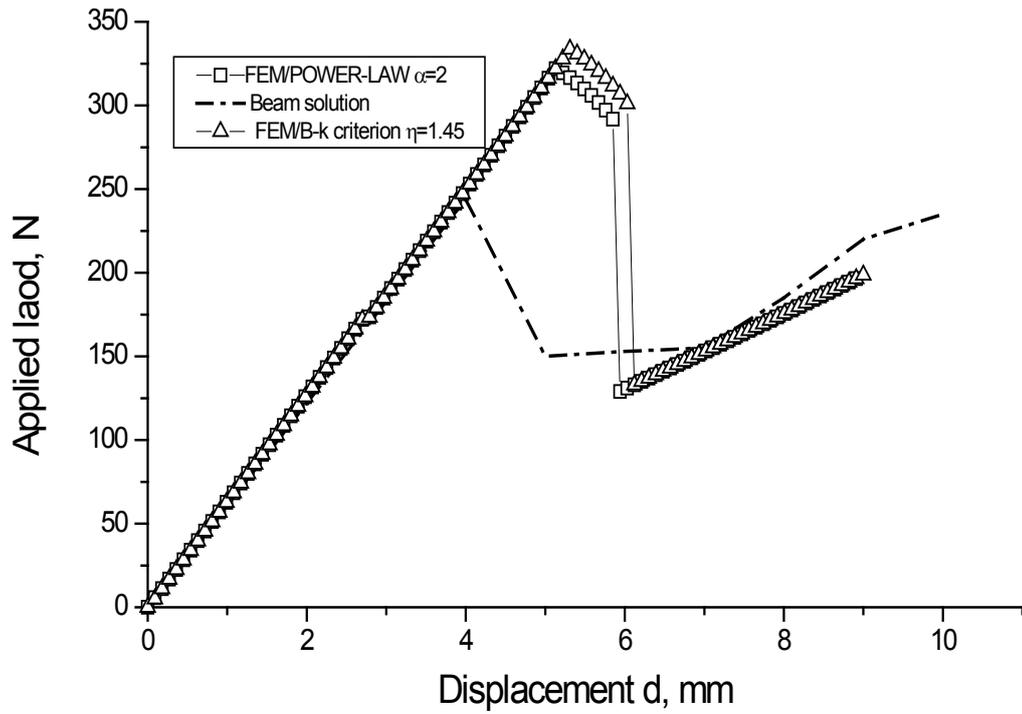


Figure 8. Load-deflection response of 3ENF test

6. New Woven Laminated Composite Debonding

Hybrid composite: In addition to the components of the reference composite, a natural load: the date core pellet was built-in thus forming the hybrid composite. The date cores powder incorporation has an increase effect of the mechanical characteristics giving to the hybrid composite a better behaviour and reducing certain types of degradation like delamination. Fig. 9 shows homogenous and uniform date pellets distribution in the hybrid composite. It is necessary also to notice the phenomenon of date cores granulate intermingling around the fabric wicks. On figure 10, it appears fiber decoherence of glass fiber and illustrates the fiber rupture. We can see a consolidation due to date cores granulates. The fibers appear well intermingled. This phenomenon gives a better mechanical behaviour to the hybrid composite.

Numerical simulations were carried out in end-notched flexure (3ENF) tests to detect initiation and growth of delamination in the hybrid composite. The length of specimen modelled is 60 mm, its width is 22 mm, and composed of two 1.65 mm thick plies. The thickness of the interface is taken equal to 1/5 of specimen thickness. Figure 11 shows the numerical predictions and experimental data for the 3ENF tests of the woven laminated composite. As it was mentioned earlier in this paper, the objective of the present work is to develop a delamination model that can predict delamination growth in new woven laminated composite for orthopaedic use. The reference composite consists of an organic matrix containing methyl methacrylate and of a woven reinforcement including: a

reinforcing glass fiber and fabric perlon having an absorbing role. The textile reinforcement made up of several folds reinforcing laid according to the orientation [90/45₂/0] [48].

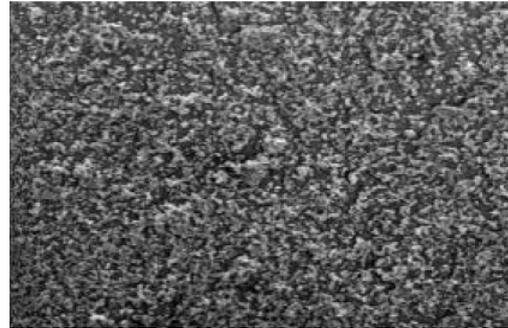


Figure 9. Micrograph of a polished area of one intermingled with fiber glass twill and granulated of date cores observed to MEB during the development of the hybrid composite [48]

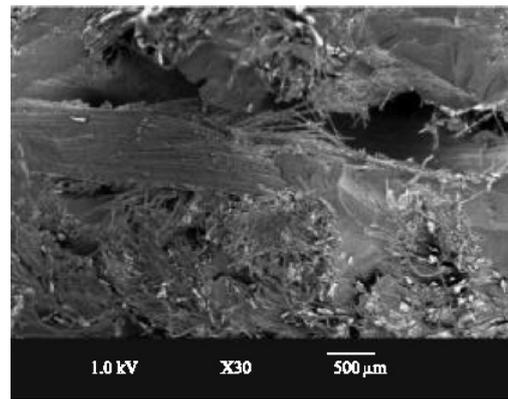


Figure 10. Fracture topography of hybrid composite [48]

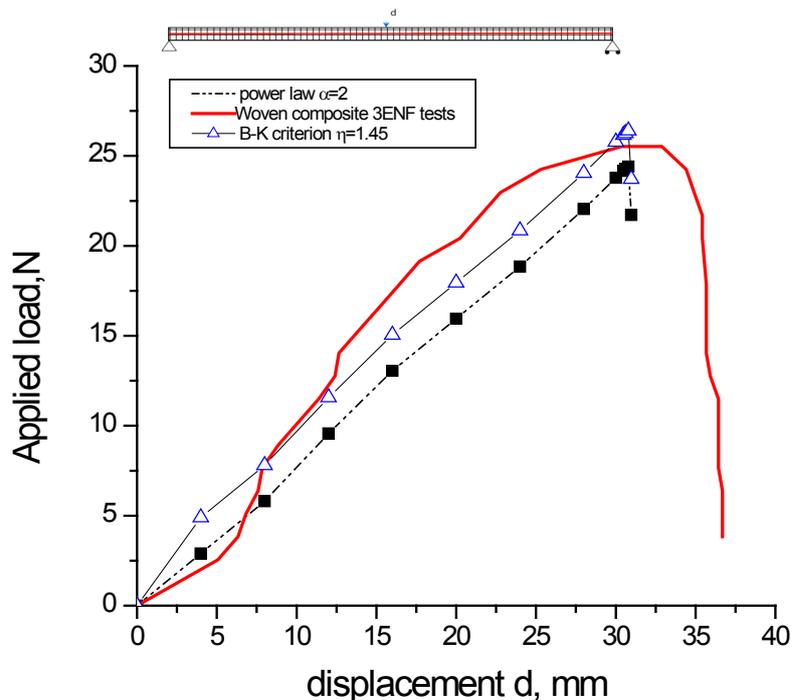


Figure 11. Experimental and predicted curves of the woven laminated composite 3ENF tests

Table 1. The geometry of test specimen⁴⁷

Specimen length (mm)	100
H (mm) - ply thickness	1.50
B (mm) –Specimen width	10

Table 2. The proprieties of the composite graphite/Epoxy ⁴⁷

E_{11}	$E_{22} = E_{33}$	$\nu_{12} = \nu_{13}$		$G_{12} = G_{13}$
150GPa	11 GPa	0.25	0.45	6.0 MPa
G_{IC}	G_{IIC}	$\sigma_{33,m}$	$\sigma_{13,m} = \sigma_{23,m}$	k
0.268 N/mm	1.45 N/m	30 MPa	40MPa	10^6 N/mm

Table 3. The proprieties of the woven laminated composite

E_{11}	$E_{22} = E_{33}$	$\nu_{12} = \nu_{13}$		$G_{12} = G_{13}$
1.1GPa	-	0.25	-	-
G_{IC}	G_{IIC}	$\sigma_{33,m}$	$\sigma_{13,m} = \sigma_{23,m}$	k
-	0.082 N/m	-	4MPa	248N/mm

7. Conclusions

The version of this template is V1. Most of the formatting instructions in this document have been compiled by SAP Productions. SAP Productions offers A4 templates for Microsoft Word. SAP Productions has tried its best efforts to ensure that the templates have the same appearance a method for the simulation of progressive delamination based on spring elements was presented. Spring elements *COMBIN 14* are placed between layers of solid elements that open and shear in response to the loading situation. The onset of damage and the growth of can be simulated without previous knowledge about the location, the size, or the direction of propagation of the delamination. A softening law for mixed-mode delamination that can be applied to any interaction criterion was proposed. The constitutive equation proposed uses thermodynamic conjugate forces variables, to track the damage at the interface under general loading conditions. The material properties required to define the element constitutive equation are the interlaminar fracture toughnesses, the penalty stiffness, and the strengths. The ENF test examples were presented that tests the accuracy of the method. Simulations of the ENF test represent cases of single-mode delamination. The examples analyzed are in good agreement with the analytical results and they indicate that the proposed formulation can predict the strength of composite structures that exhibit progressive delamination. The formulation can be extended to composite structures without any pre-existing defect. The new woven laminated composite for orthotropic use, debond problem was used as

complex structure to test the capabilities of the method. The examples analyzed indicate that the method of interface considered as uniform distribution of springs is capable of accurately predicting delamination growth. This study shows also the positive effect of date cores powder incorporation on the mechanical characteristics giving to the new hybrid composite a better resistance and reducing delamination.

REFERENCES

- [1] A. C. Garg, A damage model in composite structures, Engng. Fracture Mech., 1998. vol. 29.
- [2] O. O. Ochoa. and ReddyJ.N, Finite elements analysis of composites laminates, Kluwer, 1992.
- [3] S. Jain and D.C.H. Yang, Effects of federate and chisel edge on delamination in composite drilling. Processing and Manufacturing of Composite Materials, ASME PED 49. 1991. Vol. 27.
- [4] D. Engrand, Boundary layer approach to the Calculation of Transverse stresses along the Free Edge of a Symmetric Laminated Plate of Arbitrary Width under in Plane Loading, *Composites Structures*, 1981.
- [5] H. Dumontet., Study of Boundary layer Problem in Elastic composite Materials, *M²AN*, 20, 1986.
- [6] R. Y. Kim. and S.R. Sony, Delamination of Composite Laminates Stimulated by interlaminar Shear, *ASTM-STP* 893, 1986

- [7] A.S.D. Wang, M. Slomania, and R.B. Bucinell, Delamination Crack Growth in Composite Laminates. Delamination and Debonding of Materials, JOHNSON W.S (Ed.), ASTMSTP, 1985.
- [8] A.S.D. WANG, Fracture Analysis of Interlaminar cracking, Interlaminar response of Composite Materials, *Composite materials Series*, PAGANO N.J. (Ed.) Elsevier, 1989.
- [9] D.B. Davidson., R. Krüger and M. König. Effect of stacking sequence on energy release rate distribution in multi-directional DCB and ENF specimens. Eng. Fract. Mech., 1996. vol. 55-4.
- [10] T. K. O'brien, Mixed-mode strain-energy-release rate effects on edge delamination of composites, In: damage in composite Materials, ASTM STP 836, ASTM, Philadelphia, PA, 1984.
- [11] M.Fremond. Adhérence des solides, J. Mécan. Théor. Appl. 6, 1987.
- [12] M. Fremond. Contact with Adhesion. In Topic in Nonsmooth Mechanics, Birkhäuser, 1988.
- [13] N.Point, Approche Mathématiques de problèmes à frontières libres: Application à des exemples physiques, Thèse de doctorat d'Etat Es-Sciences Mathématiques de l'Université Paris XIII, 1989.
- [14] J. M. Truong Dinh Tien, Contact avec adhérence. Thèse de Doctorat de l'Université Paris VI, 1990.
- [15] L. Ascione and D, Bruno On delamination problem of two layer plates. In unilateral problems in Structural Analysis, Springer, Berlin, 1985.
- [16] A. Girmaldi and J.N, Reddy On delamination in plates: a unilateral contact approach. In Unilateral Problems in Structural Analysis, Springer, Berlin, 1985.
- [17] P.Ladevèze. A damage computational method for composite structures. Computer Structures, (1992). vol. 44.
- [18] O. Allix and P.Ladevèze, Interlaminar interface modelling for the prediction of delamination, Composites Structures, 1992.
- [19] A. Corigliano, Formulation identification and use of interface models in the numerical analysis of composite delamination, Int. J. Solids Structures, 1993. vol. 30.
- [20] J. C. Schellekens and DE R.Borst, Free edge delamination in carbon epoxy laminates: A novel numerical/experimental approach, Composites Structures, 1993. vol. 28.
- [21] Leandro José da Silva¹, Túlio Hallak Panzera, André Luis Christoforo, Luís Miguel Pereira Durão, Francisco Antonio Rocco Lahr, Numerical and Experimental Analyses of Biocomposites Reinforced with Natural Fibres, International Journal of Materials Engineering 2012, 2(4): 43-49
- [22] D. Benzerga, A. Haddi., A.Seddak. and A.Lavie, mixed-mode damage model for delamination growth applied to a new woven composite, Computational Materials Science, 2008. vol. 41, pp. 515–521
- [23] P. Ladevèze and E. Ledantec, Damage modelling of the elementary ply for laminated composites, Composites Sciences and technology, 1992. vol. 43.
- [24] O. Alix and P. Ladevèze, Interlaminar Interface modelling for Prediction of Delamination, Composites Structures, 1992. vol. 22.
- [25] F. Gruttmann and W. Wagner, On the numerical analysis of local effects, Composites structures, 1994. vol. 29.
- [26] J. C. Simo and M. S. Rifai. A, class of mixed assumed strain methods and the method of incompatible modes, Int. Num. Meth. Eng, 1990. vol. 29.
- [27] P. P. Camanho, C. G. Dávila and D.R Ambur, Numerical Simulation of Delamination Growth in Composite Materials. NASA/TP -211041, 2001.
- [28] A. F. Johnson, Holzapfel, M, Influence of delamination on impact damage in composite structures, Composites Science and Technology, Elsevier, 2005.
- [29] O. Allix, D. Lévêque, and L.. Perret, Identification and forecast of delamination in composite laminates by an interlaminar interface model, Comp. Sc. And Tech. Elsevier, 1997.
- [30] D. Weichert., A. Hachemi, Limit and shakedown analysis with decohesive effects. XXI CTAM, 15-21, Warsaw, Poland, 2004
- [31] P. Ladevèze, Inelastic strains and damage. In: Talreja R, editor, Damage mechanics of composite materials, Composite materials series, 1994. vol.9. Amsterdam: Elsevier.
- [32] Daudeville, L. Ladevèze, P. A damage mechanics tool for laminate delamination, LMT, ENS de Cachan
- [33] O. Allix, P Ladevèze, Delamination analysis by damage mechanics: some applications, LMT, ENS de Cachan.
- [34] M.F.S.F. DeMoura, J.P.M. Gonçalves, Modelling the interaction between matrix cracking and delamination in carbon-epoxy laminates under low velocity impact, Comp. Sci. and tech 64, 2004. Elsevier
- [35] O.Allix, P. Ladevèze, Corigliano, A Damage analysis of interlaminar fracture specimens, Comp. Struct. 1995.
- [36] L.Gornet, D.Lévêque, L. Perret, Modélisation, identification et simulations éléments finis des phénomènes de délaminage dans les structures composites stratifiées, Mec. Ind. 1, Ed. Scientifique et médicales Elsevier, 2000.
- [37] O. Allix, D Lévêque, and L. Perret, Identification and forecast of delamination in composite laminates by an interlaminar interface model. Comp. Sc. And Tech. Elsevier, 1997.
- [38] Z. Zou, S. R.Reid, S. Li, A continuum damage model for delamination in laminated composites, Journal of the Mechanics and Physics of Solids 51, 2003.
- [39] A. Haddi, D. Weichert, Elastic-plastic J-integral in inhomogeneous materials, Computational Materials Science, 1997. Volume 8, Issue 3, Pages 251-260
- [40] A. Haddi, D. Weichert, On the computation of the J-integral for three-dimensional geometries, Computational Materials Science, 1996. Volume 5, Issues 1-3, February-March, Pages 143-150.
- [41] N. Valoroso, L. hampaney, Mixed-mode decohesion in adhesive joints using damage mechanics and interface elements, Associazione italiana per l'Analisi delle Sollecitazioni xxxiv *Convegno Nazionale* – 14-17, Politecnico di Milano, 2005.

- [42] Noll. W Truesdell., The non-linear field theories of mechanics, In S. Flügge, editor, Handbuch der Physik Band III/3, Berlin, 1965. Springer.
- [43] G. Alfano, S. deBarros, L. Champaney and N. Valoroso, Comparison between two cohesive-zone models for the analysis of interface debonding, European Congress on Computational Methods in Applied Sciences and Engineering ECCOMAS, 2004.
- [44] P. P. Camanho, C. G. Dávila, and D. R. Ambur, Numerical Simulation of Delamination Growth, in Composite Materials, NASA/TP-211041, 2001.
- [45] ANSYS Structural and Analysis Guide. SAS IP Inc. (2001). Non-linear structural analysis : (Chapter 8).
- [46] L. Benabou. N. Benseddiq, M. Nait-Abdelaziz, Comparative analysis of damage at interfaces of composites. Composites: Part B33, Elsevier, 2002.
- [47] Y. Mi, M.A. Crisfield, Analytical Derivation of Load/Displacement Relationships for Mixed-Mode Delamination and Comparison with Finite Element Results, Imperial College, Department of Aeronautics, 1996. London.
- [48] K. Tadjine, Effet de l'incorporation de poudre de noyaux de dattes dans un composite polymère à usage biomécanique, PhD Thesis, 2007. Université d'Annaba.